

Page 10, line 19; please change "/363,362" to --362,362--.

Page 12, lines 5 and 6; please delete "Serial No. 08/(GE ATTORNEY DOCKET NO. 13DV-12148)" and insert --5,531,570--.

Page 12, line 32; after "to" please insert --the--.

IN THE CLAIMS:

Please amend the following claims as indicated.

Amended
1. (TWICE AMENDED) A gas turbine engine component comprising:

a metallic airfoil having a leading edge and a trailing edge and a pressure side and a suction side,

at least [one] a first laser shock peened surface on [at least one] a first side of said airfoil,

said laser shock peened surface extending radially along at least a portion of said leading edge and extending chordwise from said leading edge, and

C'
a first region having deep compressive residual stresses imparted by laser shock peening (LSP) extending into said airfoil from said laser shock peened surface wherein said deep compressive residual stresses extend from said laser shocked peened surface to a depth in a range of about 20-50 mils into said region.

Amended
2. (AMENDED) A component as claimed in claim 1 further comprising:

[a] said first laser shock peened surface located along said pressure side of said leading edge, [and

C2
a first region having deep compressive residual stresses imparted by laser shock peening (LSP) extending into said airfoil from said first laser shock peened surface,]

a second laser shock peened surface located along said suction side of said leading edge and extending radially along at least a portion of said leading edge and extending